

Isolated Synthetic Jet in Crossflow: Experimental Protocols for a Validation Dataset

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An overview of the data acquisition, reduction, and uncertainty estimates of experimental measurements made of the flowfield created by the interaction of an isolated synthetic jet and a turbulent boundary layer is presented. The experimental measurements were undertaken to serve as the second of three computational fluid dynamics validation databases for active flow control. The validation databases were first presented at the NASA Langley Research Center Workshop on CFD Validation of Synthetic Jets and Turbulent Separation Control (CFDVAL2004) in March 2004. Detailed measurements were made to document the boundary conditions for the flow and also for the phase-averaged flowfield itself. Three component laser-Doppler velocimetry, 2-D particle image velocimetry, and stereo particle image velocimetry were used to document the phase-averaged velocity field and the turbulent stresses.

I. Introduction

THE “active” in active flow control (AFC) implies that an actuator is attempting to exert control over a flow in a time-dependent sense. By varying the actuator output, or turning it completely on and off, the flow can be tailored in different ways at different times. In fact, the output of the actuator itself is, more than likely, an unsteady flow, exploiting the additional dynamics of startup to maximize the vortical impulse that the actuator can deliver to the flow. Moreover, the underlying flow that we wish to control is very often an unsteady, or time-dependent, flow. Flows with separated regions, flows with noise-producing free shear layers, flows for which we wish to enhance mixing, all of these are dominated by structures that exist only in a time-dependent sense. The design of future AFC actuators and actuator configurations, for both laboratory use and actual flight vehicle use, would be greatly enhanced by the availability of computational tools capable of predicting the unsteady flows typical of an AFC configuration. Advances in computational fluid dynamics (CFD) over the last decade have indicated that such a tool is conceivable in the near future, but one of the principle impediments in its development is the lack of high-quality experiments to validate such a tool against [1]. This is not to say that there are not high-quality experiments documented in the open literature, but to say that the type of details that need to be documented for the experiment to be useful as a validation experiment are not typically documented in the open literature. This is especially true when it comes to boundary conditions.

To address this lack of validation data, specifically with respect to AFC, a set of three validation experiments were conducted at NASA Langley. Each of the three experiments were designed with close collaboration between the experimentalists and the local CFD community to define the geometries used in the three cases, the actuator details, and the type and location of the data to be acquired. The three experiments form the validation databases for the NASA Langley Research Center Workshop on CFD Validation of Synthetic Jets and Turbulent Separation Control. The workshop was conducted in March 2004 and was jointly sponsored by the U.S. Air Force

Office of Scientific Research (AFOSR), in association with the European Research Community On Flow, Turbulence And Combustion (ERCOTAC), the International Association of Hydraulic Engineering and Research (IAHR), QNET-CFD, and the National Institute of Aerospace (NIA). An overview of the results of the workshop was presented by Rumsey et al. [2]. The goal of each of the validation experiments was to provide the most accurate description of each of the flowfields as possible. To successfully accomplish this goal, multiple instrument systems were used, allowing each to make its own contribution to documenting the velocity field as unambiguously as possible.

Each of the three cases involved some flowfield, or portion of the flowfield, being influenced by a synthetic jet actuator, also referred to as a synthetic jet or as a zero-mass-flux jet. The synthetic jet actuator was chosen as a representative actuator due to its widespread use in AFC work to date. Their popularity extends from the fact that they can be fabricated to a small size, they operate with zero net mass, that is, they synthesize a jetlike flowfield without being supplied by an external source of fluid, and they generate the type of disturbances required for AFC. The flowfield created by the synthetic jet, while zero-net mass, imparts a finite vortical and momentum impulse to the fluid in order to synthesize a jetlike flow in the mean. A synthetic jet consists of a closed cavity, connected to the outer flowfield by an orifice. Typically, one of the walls of the closed cavity is an active element that is able to change the volume of the cavity. Thus in a synthetic jet, the flow through the orifice and out into the outer flowfield alternates between an exhaust and a suction cycle, driven by the contraction and expansion of a cavity internal to the actuator. During the exhaust cycle, a vortex ring is formed and moves away from the orifice under the action of self-induction. The cyclic nature of the cavity expansion and contraction causes a train of vortices to be formed, and this vortex train yields, in the mean, velocity profiles similar to that of a turbulent jet [3,4]. In the open literature on AFC, one can find the synthetic jet being applied to achieve flow control on a circular cylinder [5–7], airfoils [8–11], and in internal flows [12,13].

The three cases for the workshop were selected and ordered in increasing levels of complexity with respect to the flow physics involved. It is expected that for a flow solver to be useful as a design tool it needs to be able to simulate each of these cases with some degree of fidelity. The first of the three, case 1, was an isolated synthetic jet issuing into a quiescent environment [14]. This case highlights the development of the vortex ring as the actuator goes through its exhaust cycle, the entrainment of the ambient fluid by the ring structure, and the movement of the ring away from the wall. The breakdown of the ring and the interaction between each ring and the previously formed ring are also key features of the flow that were documented. The second case, case 2, builds upon the isolated

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synthetic jet of case 1 by adding a crossflow, with no streamwise pressure gradient, for the synthetic jet to interact with. Here, the vortex ring formation is delayed and distorted by the crossflow. The jet starts off initially being issued normally into the crossflow but will turn to be aligned with the crossflow as the jet and the crossflow interact away from the orifice. The last case, case 3, involves the flow over a wall-mounted “hump” [15]. The flow here is characterized by a turbulent boundary layer flowing over a wall-mounted disturbance, the hump, and adds the complexities of streamwise pressure gradients and a fixed separation location. Case 3 was examined in three conditions, the baseline flow, steady suction, and control of separation by synthetic jet actuation. It should be noted that although the cases are ordered in increasing flow physics complexity, case 2 stands out as being the one case that cannot be approximated as being two dimensional, it is inherently three dimensional. Therefore, case 2 may actually be the most challenging case from a computational standpoint.

Formally, case 2 examines the interaction of a single, isolated, synthetic jet and a fully turbulent zero-pressure gradient boundary layer at a single crossflow Mach number and a single jet exit velocity. No parametric studies were performed, just a single test condition. The resulting flow has many of the characteristics that need to be modeled with fidelity if the results of similar calculations are to serve as the basis for research and design with active flow control devices. These include the turbulence in the boundary layer, the time evolution of the large vortical structure emanating from the jet orifice and its subsequent interaction with and distortion by the boundary layer turbulence, and the effect of the suction cycle on the boundary layer flow.

The current experimental setup is conceptually similar to that documented by Schaeffler [16]. The concept of studying a synthetic jet in crossflow follows from the extensive analysis of the interaction of a steady jet with a crossflow, both experimentally and computationally. The steady jet injected normal to a crossflow establishes the baseline for these types of flow. In the steady case, the jet is observed to reorient, as it develops in the streamwise direction, from an orientation normal to the crossflow to being aligned with the crossflow. During this process, the jet column transforms into a pair of counter-rotating vortices [17]. Additionally, there are other vortical structures near the jet exit that exist due to the interaction of the crossflow boundary layer and the orifice [18]. The vorticity contained in the crossflow boundary layer is reorganized by its interaction with the jet, forming a number of structures in the wake of the jet, ranging from horseshoe vortexlike structures to vortices that connect the jet to the boundary layer along the wall [19]. Adding an unsteady component to the jet, by some modulation of the jet velocity, increases the penetration of the jet into the crossflow [20] and introduces some nomenclature for describing the degree of unsteadiness in the jet. Modulation on top of a mean flow creates the pulsed jet [20], turning the jet completely on and off yields a fully modulated pulsed jet [21], decreasing the frequency causes the formation of discrete vortex rings [22] with no mean jet flow, and finally increasing the frequency and adding a suction cycle, to achieve zero-net-mass flux, yields a synthetic jet [23–27].

To document the flowfield of case 2, with all of its richness, several measurement techniques were used. The upstream inflow velocity profile and several phase-averaged velocity profiles were measured with a 3-component laser-Doppler velocimetry system. Phase-averaged velocity field measurements were made with both 2-D digital particle image velocimetry and stereo digital particle image velocimetry. Surface pressure measurements were made using an electronically scanned pressure system. Dynamic measurements were made of the pressure within the actuator cavity. The actuator cavity temperature and the displacement of the actuator diaphragm were also documented. A complete list of the data comprising the database for case 2 is listed in Sec. IV. The types of data acquired illustrate that it was a goal of this experiment to describe the boundary conditions as completely as possible. The upstream inflow velocity profile provides a starting point for the crossflow, whereas the diaphragm displacement drives the flow inside the cavity that generates the synthetic jet. All of the data is available online, at the

Internet address given in the Acknowledgments, and the reader is encouraged to download the data. The focus of this paper will be more on the data acquisition and reduction of this data than is typically expected. A clear understanding of how the experimental data was acquired and how it was reduced is critical in order to compare CFD results with the experimental results.

II. Description of the Facility and Synthetic Jet Actuator

This research effort was conducted in the NASA Langley 15-inch Low Speed Wind Tunnel of the Flow Physics and Control Branch. This tunnel is a closed-return atmospheric facility dedicated to basic flow physics research efforts. The tunnel has a maximum speed of 115 ft/s and a turbulence intensity of less than 0.13%. The tunnel medium is air at sea level conditions. During this research effort, the tunnel was equipped with a suspended flat plate model that acts as a splitter plate and facilitates the installation of the actuator within the plate model. The plate features an elliptical leading edge. Immediately downstream of the transition from the leading edge to the plate, there is a grit strip to trip the boundary layer of the plate and also grit strips at corresponding locations on the tunnel walls. The ceiling of this tunnel is adjustable at several locations down the length of the tunnel. The ceiling height was adjusted to create a zero-pressure gradient over the plate starting from a station 36.5 in. (927 mm) downstream from the transition from the elliptical leading edge to the flat plate to a station 70.5 in. (1791 mm) downstream. The elliptical leading edge section of the splitter plate is 12.0 in. in length. The centerline of the jet was located 50.5 in. (1283 mm) downstream of the flat plate start (62.5 in. from the leading edge). The diameter of the synthetic jet used here was 0.25 in. (6.4 mm). Therefore in terms of jet diameters, the region of the zero-pressure gradient extended 56 jet diameters upstream of the jet centerline and 79 jet diameters downstream, as shown in Fig. 1. These pressures were measured via a row of static pressure taps which are installed in the splitter plate, along the centerline of the tunnel, with a total of 15 taps upstream and 15 taps downstream of the jet centerline. A set of spanwise pressure taps, 16 in total, were located 8 jet diameters upstream of the jet centerline and another set of spanwise taps, also 16 in total, were located 8 jet diameters downstream. All of these taps were connected via short lengths of flexible Tygon tubing to two electronically scanned pressure (ESP) transducers, each with a full scale range of ± 10 in. of water. The ESP modules were mounted inside the plate model and the measured pressures are referenced to the static pressure of the tunnel, as observed by a pitot-static tube upstream of the splitter plate. The modules were automatically calibrated at regular intervals during a given test day. The errors in the coefficient of pressure measurements are estimated to be $C_p \pm 0.002$. The

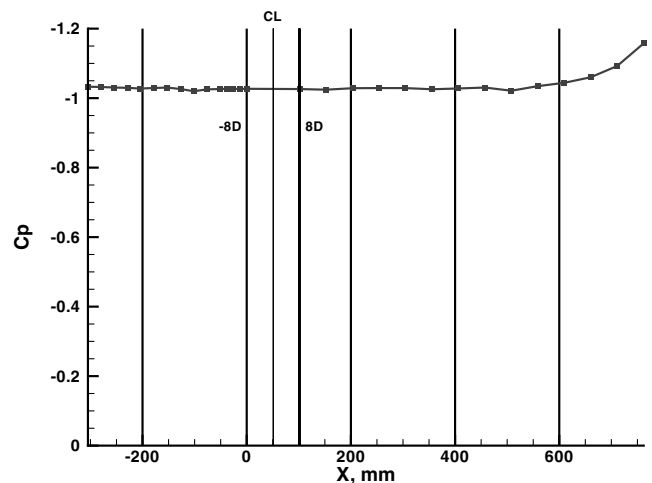


Fig. 1 Pressure distribution down the centerline of the splitter plate measured in the model coordinate system (MCS). CL marks the location of the jet centerline in the MCS.

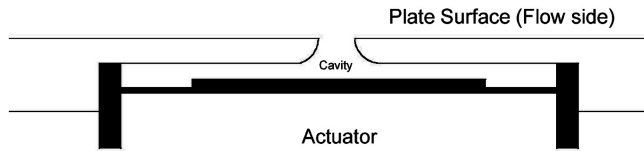


Fig. 2 Schematic of the synthetic jet actuator.

pressure data were used to calculate the freestream velocity over the plate and to verify that the pressure gradient over the plate, in the area of interest, was zero. Subsequent LDV measurements indicate that the variation in the freestream velocity over the plate, as indicated by the turbulence intensity, is small at 0.88%. Flow visualization conducted during previous studies in this facility on a similar setup suggests that the flow is two dimensional in the zero-pressure gradient region of the plate, with the exception of the region close to the side walls of the tunnel [13]. The two dimensionality for the current setup was confirmed by the surface pressure data taken via the spanwise pressure taps, which indicated less than 1% pressure variation in the spanwise direction.

The synthetic jet used in this research effort featured a circular orifice, 0.25 in. (6.35 mm) in diameter, the exact geometry of which is part of the database. The throat of the orifice is smoothly tapered from a diameter of 0.60 in. (15.2 mm) on the inside cavity wall to the 0.25 in. diameter exit dimension. The upper cavity wall is 0.170 in. (4.32 mm) thick. The actuator design is shown schematically in Fig. 2. The active element of the actuator was the wall of the cavity opposite the exit of the jet, the bottom wall in Fig. 2. This wall was displaced electromechanically by the application of a sinusoidal voltage. To document the in situ performance of the actuator, the displacement of the actuator diaphragm was recorded, as was the cavity pressure and temperature. Two dynamic pressure transducers were installed in the interior of the cavity, an absolute pressure transducer and a differential pressure transducer. The differential pressure transducer had a range of ± 5.0 psi and was referenced to the static pressure of the tunnel upstream of the splitter plate. A thermocouple was also located in the cavity to measure the cavity temperature.

III. Measurement Systems

It was a stated goal of the three validation experiments to measure their respective flowfields as unambiguously as possible. This required the use of redundant measurements of the velocity field. The velocity measurements to support case 2 were carried out using a 3-component laser-Doppler velocimetry (LDV) system, a stereo digital particle image velocimetry (SDPIV) system, and a 2-D digital particle image velocimetry (2D-PIV) system. Each of the three measurement systems share some similar features. None of the three systems actually measure fluid velocity. Each actually measures the velocity of tracer particles that are added to the flow. The LDV system can be regarded as a “point” measurement; the measurement is made within a $100\ \mu\text{m}$ sphere and is based on the velocity of a single particle. PIV systems are regarded as an area measurement; the measurement is made based on the displacement of multiple particles within a square area, typically $400\text{--}500\ \mu\text{m}$ on each side. The reader interested in further details is referred to one of the excellent review articles or monographs available on LDV [28,29] and PIV [30–32] for more information.

The LDV system used here was an orthogonal, crossed-fringe, fiber optic probe configuration, with the probes mounted 90 deg from one another. The 514.5, 496.5, and 476.5 nm wavelengths from an argon-ion laser were used to measure the spanwise (V), vertical (W), and streamwise (U) velocity components, respectively, within the tunnel. Bragg cells were used to provide directional measurement capability in all three velocity components. Both fiber optic probes used 750-mm focal length lenses, which along with an input beam diameter of 3–4 mm for each of the six beams, generated a sample volume common to all three pairs that was calculated to be approximately $100\ \mu\text{m}$ in diameter and spherical in shape. All velocity measurements were made in coincidence mode.

Coincidence mode requires that a validated velocity signal be made independently in each of the three components within a very short time window, guaranteeing that the velocity signals come from a single particle traversing the measuring volume. The use of coincidence mode in the present research effort is required to obtain estimates for each of the six components of the Reynolds stress tensor.

Additionally, the actuator drive and sync signals were acquired with each trigger and stored if the velocity data were validated as being in coincidence. Consideration of the actuator drive and sync signals allows the LDV data, which are acquired at random intervals, to be reconstructed as a function of phase. The fiber optic probes were mounted on a three-axis traversing system, which used several motorized slides that provided 1 m of travel in each of the three axes, with $1\ \mu\text{m}$ resolution in each direction. For the LDV portion of the experiment, the flow was seeded with monodisperse, $0.86\ \mu\text{m}$ polystyrene latex (PSL) microspheres. Since the PSL particles are created via a chemical reaction, which allows their diameters to be tightly controlled, the observed variance of the particle sizes is small, $0.058\ \mu\text{m}^2$. The seed particles were suspended in 100-proof alcohol and atomized by a six-jet atomizer. The atomized particles were then introduced into the settling chamber of the tunnel. The particles were fabricated at NASA Langley using the technique described by Nichols [33].

The two PIV configurations employed a standard commercial system which incorporated the timing control, image acquisition, data management and postprocessing of both the 2D-PIV and the SDPIV data into a single computer-based system. The light source used was a pulsed, frequency-doubled 300 mJ Nd-YAG laser operating at 10 Hz. For all the PIV data, the flow within the tunnel was seeded with mineral oil atomized by a commercial six-jet atomizer to a mean particle size of $3.6\ \mu\text{m}$, with an observed variance in particle size of $1.05\ \mu\text{m}^2$. Both of the seed particle sizings were experimentally verified by a commercial aerodynamic particle sizer.

The seed was injected into the flow in the tunnel settling chamber. Standard sheet-forming optics were used and the cameras were positioned on opposite sides of the tunnel for the SDPIV data. Also for the SDPIV, the camera mounts allow for the required tilting of the lens, relative to the camera body, to satisfy the Scheimpflug condition. The light sheet orientation for the SDPIV data was in the spanwise direction and in the streamwise direction for the 2D-PIV data. The position of the sheet in the spanwise direction was controlled by a prism mounted on the same traversing system used by the LDV system. The light sheet thickness was typically 1.5 mm thick. The resolution of each camera was 1008×1018 pixels and the typical field of view was $40\ \text{mm} \times 40\ \text{mm}$ for the SDPIV and $18\ \text{mm} \times 18\ \text{mm}$ for the 2D-PIV. An interrogation window size of 32×32 pixels with 50% overlap was used for the PIV data. The PIV data were processed with a standard PIV interrogation technique by performing cross correlations on equal-sized windows in both images.

The image acquisition and lasers for both PIV systems were synchronized with the drive signal of the actuator so that images could be acquired at a specific time delay after the beginning of an actuator drive cycle. A specific delay for the laser pulsing and image acquisition corresponds to a specific phase of the actuator drive cycle. The same sync signal that was used by the LDV system was used as the trigger for the PIV signal. Within the PIV software, the timing delay was varied in order to acquire data at 36 specific phases of the synthetic jet actuator drive signal, equally spaced over both the exhaust and the suction cycles. For the 2D-PIV data, the flowfield over the entire drive signal was recorded as 14,400 instantaneous velocity fields, 400 velocity field measurements at each of 36 specific delays, which results in the measurement of the velocity field every 10 deg of phase. The 400 image pairs at each phase were acquired over a period of about 3 min. The image pairs for each phase were then processed using a standard PIV cross-correlation data reduction technique and then averaged to give the phase-averaged velocity field for that particular phase. 300 image pairs were used for each phase for the SDPIV data.

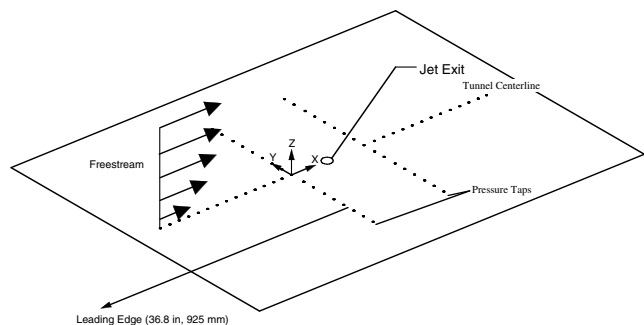


Fig. 3 Schematic of the model coordinate system.

IV. Boundary Conditions

The boundary conditions for case 2 consist of the upstream, or inflow, boundary condition, the geometry of the model and the tunnel itself, and the boundary conditions associated with the actuator, that is, the diaphragm displacement, cavity pressure, and cavity temperature as a function of phase. The tunnel test condition was a Mach number in the test section over the plate of $M = 0.10$. The tunnel dimensions at the test section are 15.0 in. (381 mm) wide by 9.8 in. (249 mm) high (distance from the splitter plate to the top wall). The orifice diameter was 0.25 ± 0.005 in. (6.35 ± 0.13 mm). Measurements presented herein are presented with reference to the model coordinate system. The model coordinate system is shown schematically in Fig. 3 and its origin was established where the first line of spanwise pressure taps intersects the line of centerline pressure taps. This location is 60.5 in. (1537 mm) from the leading edge of the plate. In the model coordinate system, the centerline of the jet orifice is located at 2.00 ± 0.020 in. (50.8 ± 0.50 mm). Moving from the origin of the model coordinate system to the jet centerline in the streamwise direction defines the X coordinate axis, the Y coordinate axis is in the spanwise direction, and the Z coordinate axis is in the vertical direction. The corresponding instantaneous velocity components will be u , v , and w . The mean velocity components will be denoted by U , V , and W and the six turbulent stresses by $\overline{u'u'}$, $\overline{v'v'}$, $\overline{w'w'}$, $\overline{u'v'}$, $\overline{u'w'}$, and $\overline{v'w'}$. From this point forward, distances in the streamwise X direction will be stated in millimeters or, at times, jet diameters from the jet centerline. The key locations are $X = 44.45$ mm ($-1D$), $X = 50.80$ mm ($0D$, CL), $X = 57.15$ mm ($1D$), $X = 63.50$ mm ($2D$), $X = 69.80$ mm ($3D$), and $X = 76.20$ mm ($4D$). All velocity components and turbulence quantities presented hereafter, unless noted otherwise, have been nondimensionalized by the freestream velocity, or the freestream velocity squared.

Although the atmospheric conditions varied on a daily basis, the conditions were never far from standard atmospheric conditions at sea level in a wind tunnel vented to the atmosphere, in a temperature-controlled room. These conditions are given in Table 1.

The upstream boundary conditions were measured at station $X = 0.0$ mm in the model coordinate system. A complete description of the state of the turbulent boundary layer was measured using the 3-component laser-Doppler velocimetry system to be described in the following section. All six components of the turbulent stress tensor were documented and presented as the upstream boundary condition. The velocity means and the turbulent stress components were compared to similar values available in the open literature [34] and found to be in good agreement, as shown for the mean profile of

Table 1 Atmospheric conditions observed for case 2

Description	Symbol	Value
Freestream Mach number	M_∞	0.10
Atmospheric pressure	P_{atm}	101,325 Pa
Freestream temperature	T_∞	23.9°C (75°F)
Freestream density	ρ_∞	1.185 kg/m ³
Freestream viscosity	μ_∞	18.4×10^{-6} kg/ms

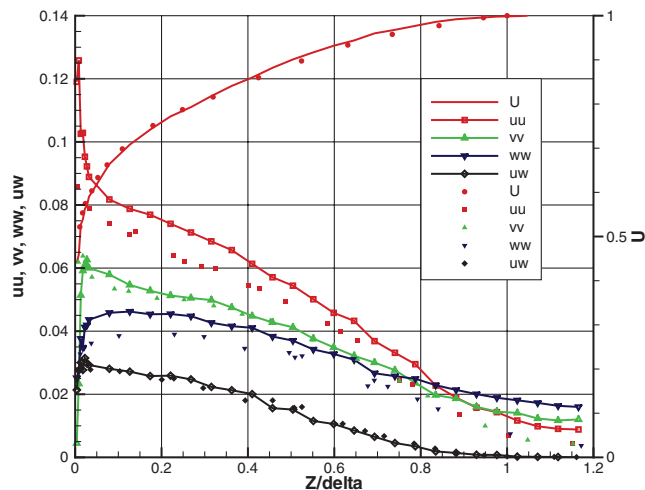


Fig. 4 Upstream boundary condition (UBC) compared to the data of Klebanoff [34]. Solid symbols without lines are the Klebanoff data. Data with lines and symbols are the UBC. All velocity data are nondimensionalized by the freestream velocity.

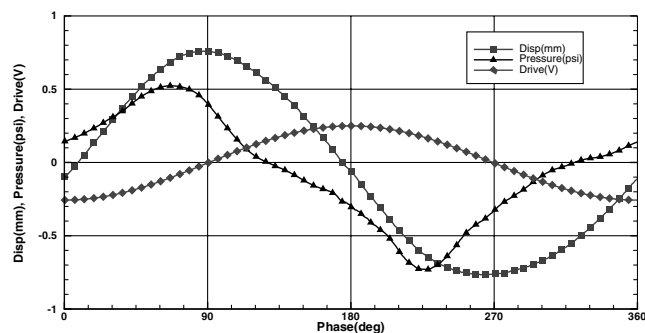


Fig. 5 Diaphragm displacement, drive signal, and cavity pressure as a function of phase for the synthetic jet actuator.

the freestream component of the velocity U and some of the turbulent stresses $\overline{u'u'}$, $\overline{v'v'}$, $\overline{w'w'}$, and $\overline{u'w'}$ in Fig. 4. The comparison here to the Klebanoff data is not to imply that the present research effort was an attempt to duplicate the experimental setup of Klebanoff. But rather, it is reflective of the use of the Klebanoff data as a “standard” zero-pressure gradient turbulent boundary layer profile by the CFD community and to illustrate the differences that can be introduced if this is used as the upstream boundary condition in lieu of the experimentally observed boundary condition.

The final components of the boundary conditions are those associated with the actuator itself, that is, the diaphragm displacement and the cavity pressure and temperature as a function of phase. The phase-averaged diaphragm displacement and cavity pressure can be seen in Fig. 5, along with the actuator drive signal. The presentation of all of the subsequent velocity measurements uses this same phase reference. The diaphragm displacement is relative to the nominal cavity depth when the tunnel is on condition. This value is 1.68 mm. The cavity temperature was observed not to be a function of phase. The cavity temperature was elevated from the tunnel temperature by 2°F (77°F).

The assemblage of the geometry description, the upstream boundary condition, the actuator boundary condition, and the velocity field data comprise the complete acquired dataset, or database, for this case. The database is specifically composed of the following items:

- 1) internal geometry definition of the actuator cavity and orifice;
- 2) upstream boundary condition. Velocity profile of U , V , W , $\overline{u'u'}$, $\overline{v'v'}$, $\overline{w'w'}$, $\overline{u'v'}$, $\overline{u'w'}$, and $\overline{v'w'}$ in the Z direction at $X = Y = 0$;
- 3) actuator displacement, pressure, and cavity temperature as a function of phase;

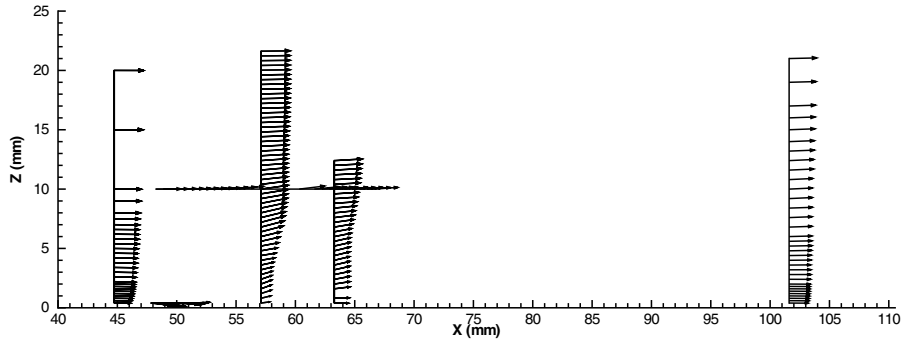


Fig. 6 Long-time average velocity profiles from the LDV data. The jet exit is located from $47 \leq X \leq 54$ and the upstream boundary condition of Fig. 4 would be located at $X = 0$.

4) jet exit velocity as a function of phase;

5) 3-component LDV velocity profiles in the Z direction at $X = 44.45$ mm (−1D), $X = 57.15$ mm (1D), $X = 63.50$ mm (2D), $X = 69.80$ mm (3D), $X = 76.20$ mm (4D), and $X = 101.60$ mm (8D);

6) 2-component PIV velocity measurements made in a streamwise plane, at $Y = 0$;

7) 3-component PIV (stereo) velocity measurements made are spanwise planes at $X = 69.80$ mm (3D), $X = 76.20$ mm (4D).

V. Data Reduction, Uncertainty, and Comparison

As discussed previously, the LDV system and the PIV system use different techniques to measure velocity. The LDV is a point measurement based on the velocity of a single particle. The PIV system is an area measurement, based on the displacement of multiple particles in a small area illuminated by a laser sheet. The two measurement systems also see the phase of the jet in different ways. The PIV system can be triggered, to a high degree of accuracy, to acquire its images at a particular time. This means that the PIV system can be triggered when the actuator drive signal is exactly at a 30 deg phase. The LDV system, in contrast, acquires data at random intervals, corresponding to whether there is a particle in the sample volume that generates a valid signal in all three components. We need to sort the LDV data by phase, after it is acquired, in order to phase average the data. This must be done by binning some range of phase angles together, in order to get enough points to generate meaningful statistics. So the LDV system provides information that is at a point in space, but “smeared” in time, whereas the PIV system is at a point in time, but smeared in space.

A. LDV Data Reduction

It is the nature of the LDV data that the velocity is randomly sampled in time. Valid velocity measurements are made only when there is a single particle in the sample volume which scatters sufficient light to generate a valid signal in each of the three velocity components. Once the data are acquired, they must be sorted based on the phase of the actuator at the time of acquisition. The LDV data system acquires, along with the coincident velocity data, data for the drive signal of the actuator and a sync signal, a square wave in phase with the drive signal. Analysis of the value of the drive signal and the value of the sync signal allows for the phase angle at the time of acquisition to be calculated. Once a phase angle is assigned to all the LDV data, the data can be sorted into bins according to phase angle. Each bin represents data with a phase angle in a certain range; the width of the bins used here was 10 deg. For example, all the data from 0 deg, inclusive, to 10 deg, exclusive, was grouped in a single bin. The outliers in the data were then eliminated based on the number of samples in the bin and Chauvenet’s criterion [35]. The data could then be averaged and the turbulence quantities calculated. These calculated values are the phase-averaged data for this bin and were assigned a phase angle in the center of the bin, in this example 5 deg. Analysis of the average phase angle in all the bins provides a value very close to the halfway point indicating that there is not a bias toward one side of the bin or the other. Once the data in each

individual bin have been phase averaged, the data from the 36 bins, or each individual phase, were averaged together to calculate the long-time average profile, the long-time mean. The long-time average velocity profiles derived from the LDV data are shown in Fig. 6.

The phase-averaged statistics, the mean velocity and turbulence quantities, for each bin of the LDV data were computed using weighted averages [28]. Specifically, the following formulations were used:

$$U = \frac{\sum u_i g_i}{\sum g_i} \quad \overline{u'u'} = \frac{\sum (u_i - U)^2 g_i}{\sum g_i}$$

$$\overline{u'v'} = \frac{\sum (u_i - U)(v_i - V) g_i}{\sum g_i} \quad V = \frac{\sum v_i g_i}{\sum g_i}$$

$$\overline{v'v'} = \frac{\sum (v_i - V)^2 g_i}{\sum g_i} \quad \overline{v'w'} = \frac{\sum (v_i - V)(w_i - W) g_i}{\sum g_i}$$

$$W = \frac{\sum w_i g_i}{\sum g_i} \quad \overline{w'w'} = \frac{\sum (w_i - W)^2 g_i}{\sum g_i}$$

$$\overline{u'w'} = \frac{\sum (u_i - U)(w_i - W) g_i}{\sum g_i}$$

When free-running weights are used, that is, $g_i = 1$, the calculated statistical quantities revert to their standard definitions, for example, U is the arithmetic mean of the u_i ’s. Use of this type of weighting scheme results in values that are an upper bound on the precision error [28].

Since all three components of velocity were acquired and the use of the coincidence mode yields a nearly spherical sample volume, the use of a weighting function based on the velocity magnitude is attractive and appropriate for use. In this case, the weighting function g_i will be the inverse of the magnitude of the velocity vector,

$$g_i = \frac{1}{\sqrt{u_i^2 + v_i^2 + w_i^2}}$$

Both the free-running and the velocity magnitude weighting functions were considered here. The mean velocity values calculated by using the free-running weights were higher than the means calculated by the velocity magnitude weights indicated that there might be some velocity bias in the data. Therefore to minimize this effect, the velocity magnitude weights were adopted for use and were applied to all of the LDV data.

Once all of the LDV data has been phase averaged, the effect of the jet on the flow can begin to be investigated. In Fig. 7, the velocity profile one diameter upstream of the jet centerline (−1D) is presented as a function of phase and with the vectors colored by the value of $\overline{u'w'}$. At this station, the effect of the jet is felt most strongly during

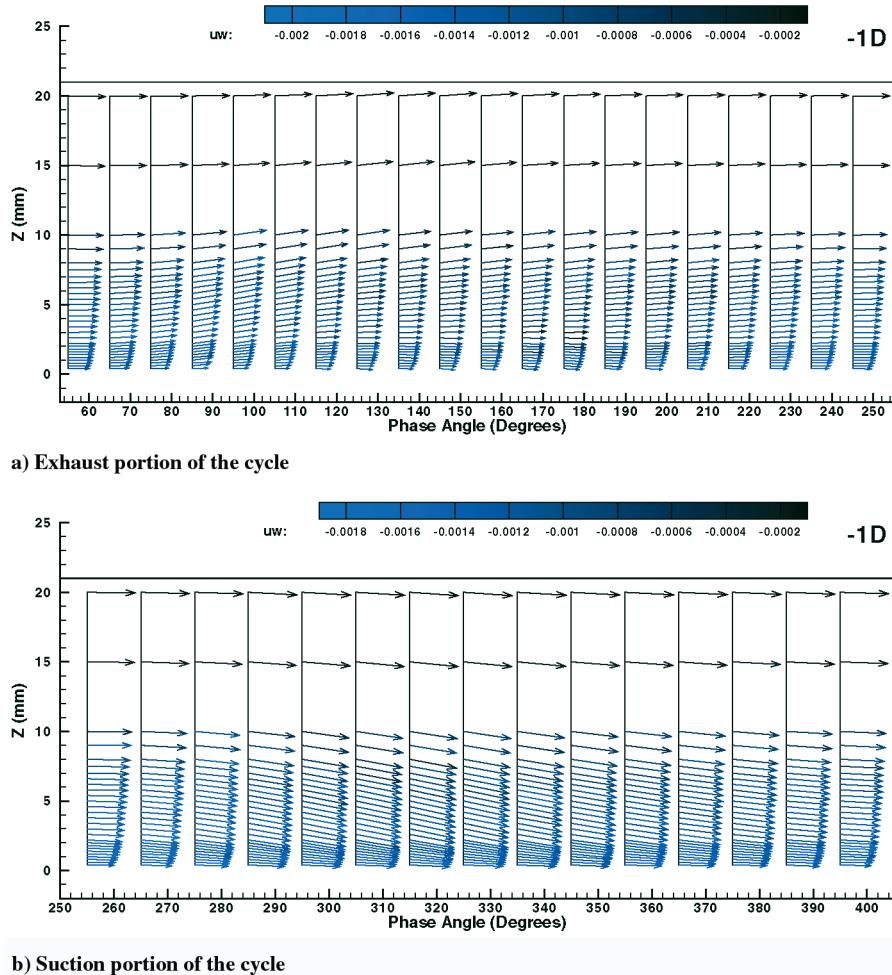


Fig. 7 Phase-averaged LDV velocity profiles at $x = 44.45$ mm ($-1D$) as a function of phase angle. The thin horizontal line at ≈ 21 mm is the height of the undisturbed boundary layer at this station.

the suction portion of the cycle. During that time, the velocity profile near the wall becomes fuller indicating that the near wall fluid is accelerated by the suction cycle of the jet. The effect of the exhaust cycle is less than the suction cycle at this station. An additional vertical component of velocity is observed throughout the profile as the fluid prepares to flow over the top of the jet, which appears as an obstruction to the oncoming flow.

At the station one diameter downstream of the jet centerline ($+1D$), the situation is quite different. Here the fluid expelled from the jet during the exhaust cycle passes through this station with a marked change in the profiles. In Fig. 8, the velocity profile one diameter downstream of the jet centerline is presented as a function of phase and again with the vectors colored by the value of $\overline{u'w'}$. The exhaust phase is marked by a large vertical component of velocity that passes through this station with corresponding large values of $\overline{u'w'}$. The suction cycle also sees a large vertical component of velocity as the fluid displaced from the wall by the jet moves back to the wall as the exhaust cycle of the jet collapses.

B. PIV Data Reduction

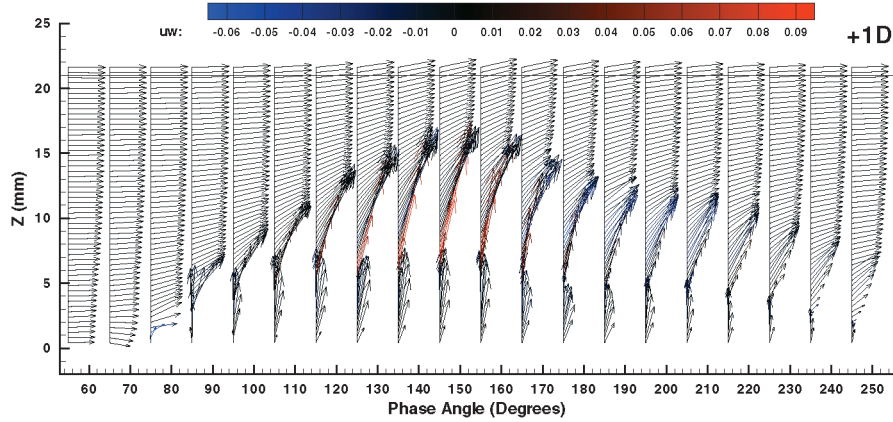
As previously described in Sec. III, the PIV system can be configured to acquire images at an exact phase of the actuator drive signal, so acquiring the data for each phase is not as problematic as it is for the LDV system. All of the image pairs are acquired and then processed using a standard PIV cross-correlation technique. The instantaneous velocity fields are validated and any missing vectors in the field are replaced by interpolation. The instantaneous velocity fields are then averaged to produce the phase-averaged velocity field and statistics for that phase. No additional processing on the instantaneous fields was done. The PIV measurements were also

taken at 10 deg intervals, so once all 36 phase averages were calculated, they were averaged together to generate the long-time mean velocity field.

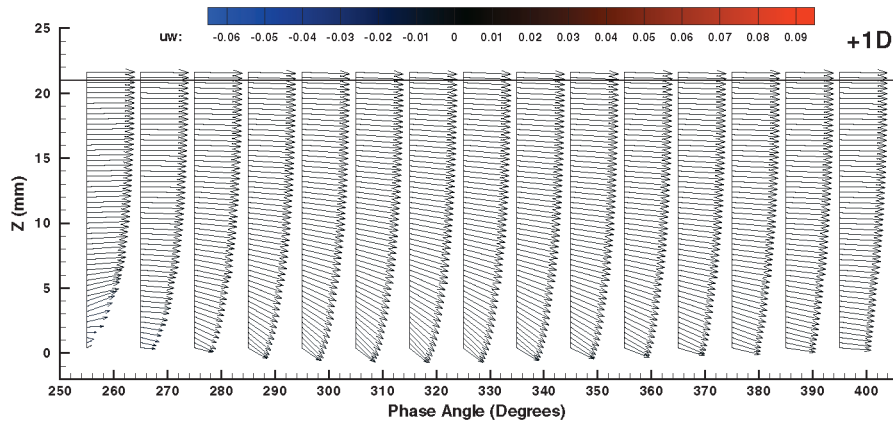
The 2D-PIV data can provide some interesting insight into the interaction of the synthetic jet and the crossflow. In Fig. 9, the velocity field at a phase of 0 deg is presented. This phase is during the suction cycle and we can see the strong influence that the jet has on the flow upstream of the actuator. All of the flow that is filling the expanding cavity has a tendency to come from upstream of the jet as shown by the instantaneous, in a phase-averaged sense, streamline pattern. In fact, the pattern shows the dividing streamline and half-saddle that form on the wall immediately downstream of the aft edge of the jet, which mark the segregation of the fluid that will tend to flow toward the actuator from the fluid that will tend to flow over the actuator.

In Fig. 10, the velocity field at a phase of 130 deg is presented. At this phase, the jet is near the peak of the exhaust cycle. The jet column can already be seen to be turning in the direction of the freestream by the edge of the frame. This height is only about half the undisturbed boundary layer thickness, illustrating that the interaction of the synthetic jet and the crossflow remains contained in the boundary layer. This was one of the requirements for the experiment.

The long-time mean of the velocity field was calculated by averaging, through one complete cycle, the individual phase-averaged velocity fields, as shown in Fig. 11 for data derived from the 2D-PIV system. We can see that in the long-term mean, the synthetic jet does appear as a steady jet which turns in the direction of the freestream close to the wall. Another long-time mean feature of the flowfield is the appearance of counter-rotating vortices in a spanwise plane. This spanwise plane, presented in Fig. 12 and measured with the SDPIV system, is 4D downstream of the jet centerline. The



a) Exhaust portion of the cycle



b) Suction portion of the cycle

Fig. 8 Phase-averaged LDV velocity profiles at $x = 57.15$ mm (+1D) as a function of phase angle. The thin horizontal line at $Z = 21$ mm is the height of the undisturbed boundary layer at this station.

development of a counter-rotating vortex pair is very similar to what occurs in the interaction of a steady jet and a crossflow.

C. Uncertainty

The mean velocities and turbulent stresses presented herein are, at best, estimates of the actual statistical values. The assignment of experimental uncertainty to each of the mean velocities and turbulent stresses is a way of providing a range on each of the measurements that has a high probability of containing the true value. This range is usually specified by a degree of confidence, or a set of odds, that the

true value does indeed lie in the given range. A confidence of 95%, or 20:1 odds, will be used here. The total uncertainty U_T takes into consideration both the bias, or instrument, error and the precision, or statistical, error [35].

The bias error U_B is composed of the errors that are present in every single measurement that an instrument makes and are not affected by the number of samples taken. They are the errors that can be minimized, but not eliminated, by careful setup, alignment, and choice of instruments. For the LDV system, one of the biggest contributors to the bias error is the accuracy with which the half-beam angle is known for all three beam pairs. [In a LDV system, each

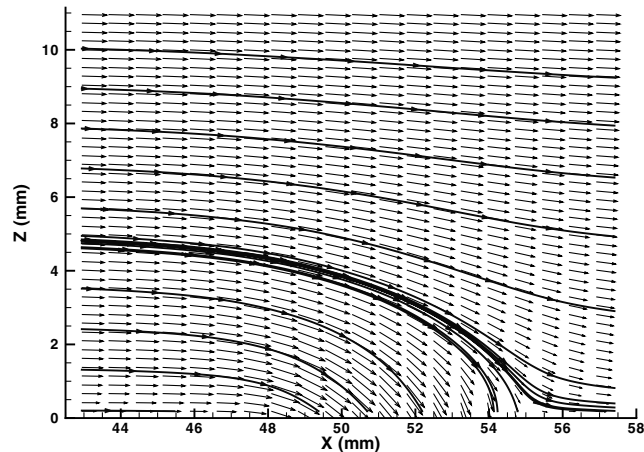


Fig. 9 PIV velocity field during the suction part of the cycle. Instantaneous streamlines indicate that a half-saddle is formed near the aft end of the jet exit during the suction cycle (0 deg phase).

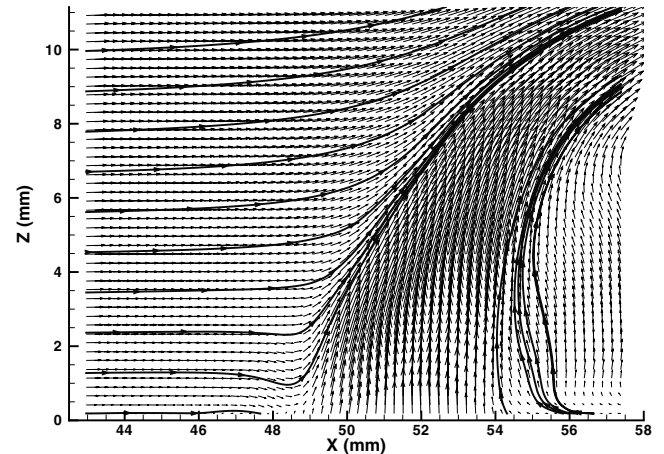


Fig. 10 PIV velocity field during the exhaust part of the cycle (130 deg phase).

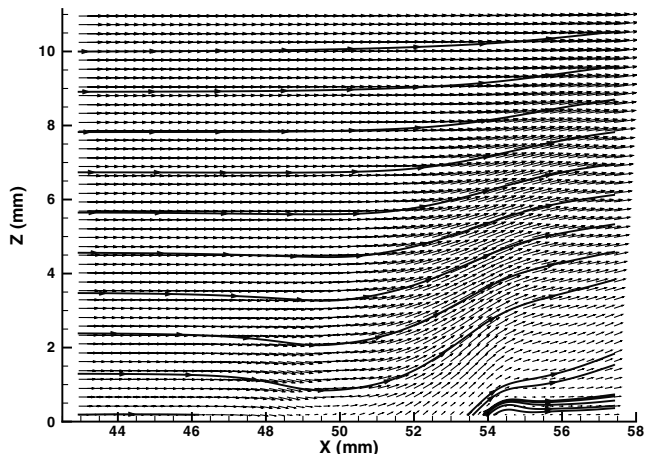


Fig. 11 Long-time mean velocity field (2D-PIV measurements).

velocity component is calculated by $v = f_d \lambda / (2 \sin \kappa)$, where f_d is the Doppler frequency and λ is the wavelength of the laser light, both of which are known with low uncertainty. Therefore, the half-beam angle κ must also be known with a low uncertainty [28.] The half-beam angle was measured through an in situ calibration, thus minimizing the uncertainty of the half-beam angle for each of the beam pairs. The bias error for the LDV data is estimated to be $\pm 0.25\%$ of the instantaneous velocity measurement $\pm 0.25\%u_i$. For the 2D-PIV system, the bias error is estimated to be $\pm 1.3\%$ of the instantaneous velocity measurement $\pm 1.3\%u_i$. Because of the nature of the equations governing the propagation of the bias errors, the bias errors in the individual velocity measurements do not propagate to the turbulent stresses [36]. Therefore, the bias error in

each of the turbulent stresses, for both measurement systems, is zero [37]. The uncertainties in the turbulent stresses are strictly precision errors.

The fact that the data taken here represent a limited number of samples of a random process requires that the effect of the sample size N on the uncertainty in our estimate of the velocity means and turbulence quantities be considered. This is captured by the precision error U_p and is related to the variances of each of the estimated velocity means and turbulent stresses. Following the work of Benedict and Gould [38], the estimator variances for each of the mean velocities and turbulence quantities were calculated as follows:

$$\begin{aligned} \sigma_{\bar{u}}^2 &= \frac{\overline{u'u'}}{N} & \sigma_{\overline{u'u'}}^2 &= \frac{\overline{u'^4} - \overline{u'u'}^2}{N} & \sigma_{\overline{u'v'}}^2 &= \frac{\overline{u'^2v'^2} - \overline{u'v'}^2}{N} \\ \sigma_{\bar{v}}^2 &= \frac{\overline{v'v'}}{N} & \sigma_{\overline{v'v'}}^2 &= \frac{\overline{v'^4} - \overline{v'v'}^2}{N} & \sigma_{\overline{u'w'}}^2 &= \frac{\overline{v'^2w'^2} - \overline{v'w'}^2}{N} \\ \sigma_{\bar{w}}^2 &= \frac{\overline{w'w'}}{N} & \sigma_{\overline{w'w'}}^2 &= \frac{\overline{w'^4} - \overline{w'w'}^2}{N} & \sigma_{\overline{u'w'}}^2 &= \frac{\overline{u'^2w'^2} - \overline{u'w'}^2}{N} \end{aligned}$$

Once the estimator variances are known, the precision error can be evaluated, at 20 to 1 odds, as being 2.0 times the square root of the variance, for example, for the streamwise velocity $U_p = 2.0\sigma_U$. Once the bias error and the precision error are known at the point in question, the total uncertainty can be calculated by the quadrature of the two.

The equations for the variances illustrate that for a given number of samples, the higher the turbulence level, the higher the uncertainty in the estimate of the mean velocity. This effect is alluded to by Uzol and Camci [39] in their analysis of the influence of turbulent intensity on the accuracy of ensemble averaged PIV measurements. During the jet exhaust, the turbulence levels, the variances seen in the

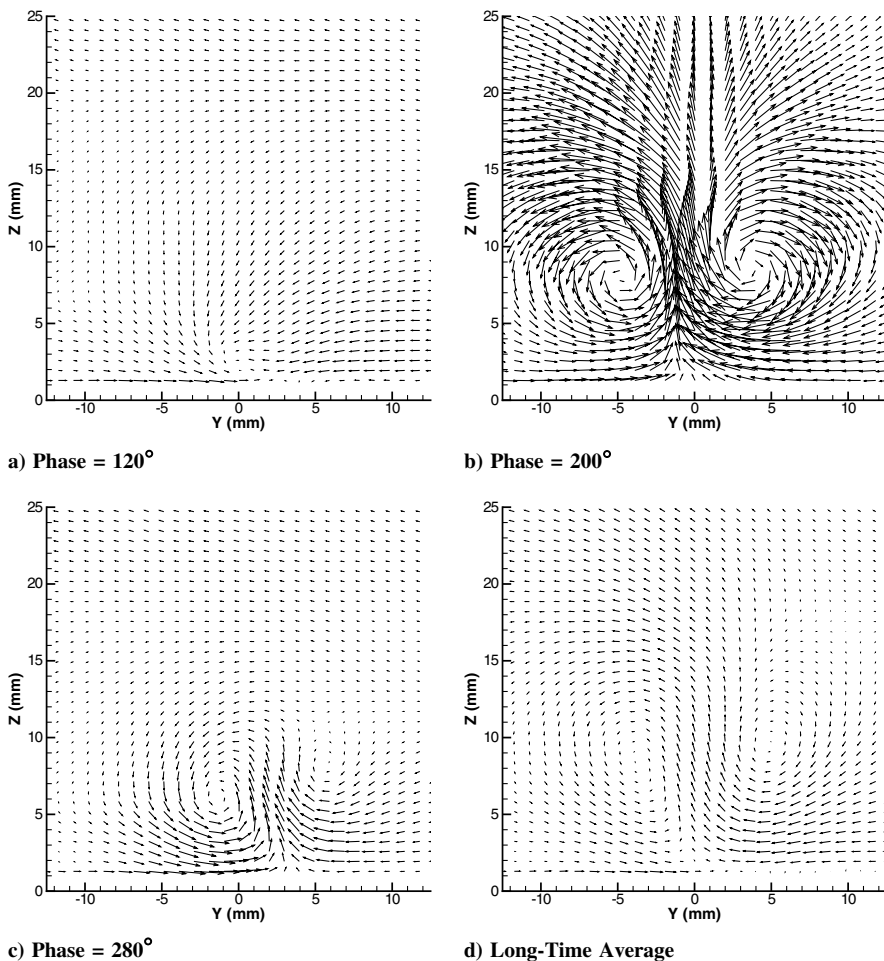


Fig. 12 Spanwise plane 4D downstream of the jet centerline (SDPIV measurements).

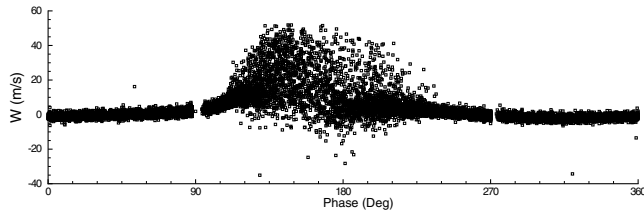


Fig. 13 LDV data for (57.1, 0.0, 10.4) after the phase data has been reconstructed but before any of the points are clipped. The gaps near 90 deg and 270 deg are artifacts of the phase reconstruction.

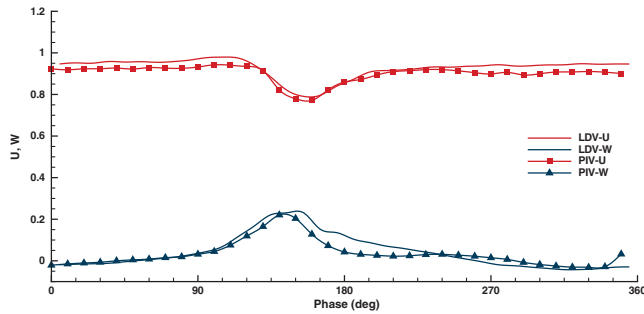


Fig. 14 Comparison of the LDV and 2D-PIV data as a function of phase at (57.1, 0.00, 14.00).

observation of the instantaneous velocities, become very high. This can be seen by looking at the raw LDV data after they have been sorted by phase. If we consider the velocity in the streamwise direction u the spread of that data at a given phase is related to the variance, which in this case is also the turbulent stress $\overline{u'u'}$. This spread can be seen in Fig. 13. For the phases not during the exhaust cycle of the jet, the spread of the data is modest and indicative of the underlying turbulence levels in the boundary layer. For phases during the exhaust cycle, the spread is much larger. The underlying turbulence and cycle-to-cycle variations in the jet formation process contribute to an increased spread of the data and calculated mean velocities for these phases will have a higher uncertainty associated with them. Therefore, in the dataset, the uncertainties are given on a point by point basis because they vary from point to point just as all the other measurements do.

D. Comparison of the LDV and the PIV data

To compare the velocity measurements made by the LDV system with those made by the PIV system, data from common locations in both datasets were examined as a function of phase. At a location 1D downstream ($X = 57.1$ mm), and 14.0 mm off the wall, we are above the disturbance of the jet. A comparison of the LDV data and the PIV data at that location, as shown in Fig. 14, shows an acceptable agreement between the two. The 2D-PIV data tend to underestimate the velocity with respect to the LDV, which may be due to the larger tracer particles used in the PIV portion of the experiment.

If we move closer to the wall at the 1D station, to 10.4 mm off the wall, the agreement worsens. This location, as shown in Fig. 15, is in the middle of the interaction of the jet and the crossflow. During the exhaust cycle, the LDV and the PIV data show different U velocity signatures. The LDV U velocity signature indicates that the streamwise velocity component U increases during the interaction with the jet and then decreases slightly from the noninteraction, or suction cycle, value ($U \approx 0.90$). The PIV U signature shows an opposite effect; the U component decreases through the interaction and then slightly increases before moving back to its value during the suction cycle. The W signature from the PIV and LDV have a closer correspondence to one another, but there are some discrepancies. The LDV W signature is slightly wider at its peak than the PIV W signature and there is a second feature in the LDV signature that is not present in the PIV signature ($180 < \text{phase} < 230$).

To determine if the discrepancy was somehow an artifact of the processing or the statistical properties of the data, several alternative

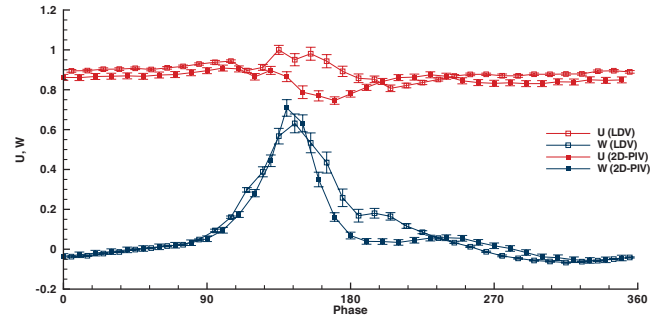


Fig. 15 Comparison of the LDV and 2D-PIV data as a function of phase at (57.1, 0.00, 10.40). Error bars are the complete uncertainty for each measurement.

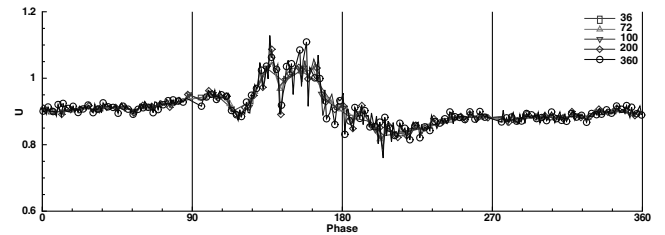


Fig. 16 Effect of increasing the number of bins used in phase averaging the LDV data.

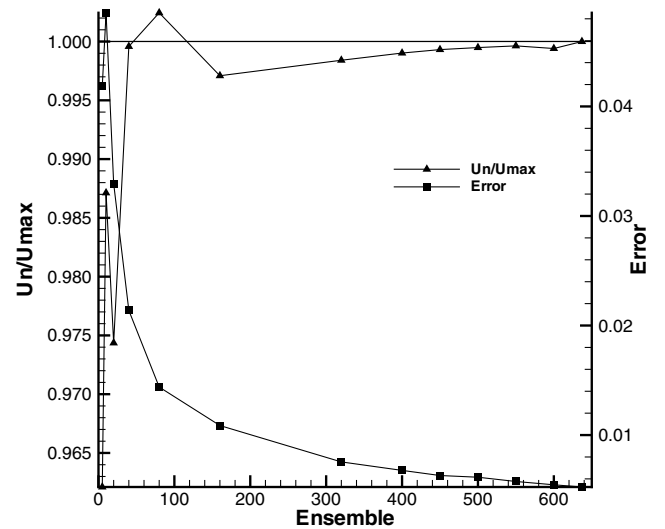


Fig. 17 Effect of sample size on the LDV phase-averaged data.

processing techniques and analyses were performed. To see if the disagreement was related to the bin size of the LDV processing, the effect of reducing the bin size was investigated. As shown in Fig. 16, the amount of noise on the U velocity signature is increased because of the reduction in the number of samples in each bin as the total number of bins is increased, but increasing the number of bins does not change the basic shape of the signature. To see if there was a sufficient number of samples to achieve a statistically stationary value for the means, the effect of the sample, or ensemble, size was investigated. As shown in Fig. 17, the ratio of the value for U based on N ensembles to the value based on the total number of ensembles available shows that the data are steadily converging toward the maximum sample value indicating that there is a sufficient number of samples [40] once that number is greater than ≈ 250 . Both the LDV and the PIV datasets are composed of more than 250 samples, implying that the means have converged in each. The PIV data were examined for evidence of “pixel locking,” which can sometimes occur when there is insufficient particle density in the PIV images [41]. As shown for a representative plane of PIV data in Fig. 18, evidence of pixel locking, peaks in the probability mass distribution

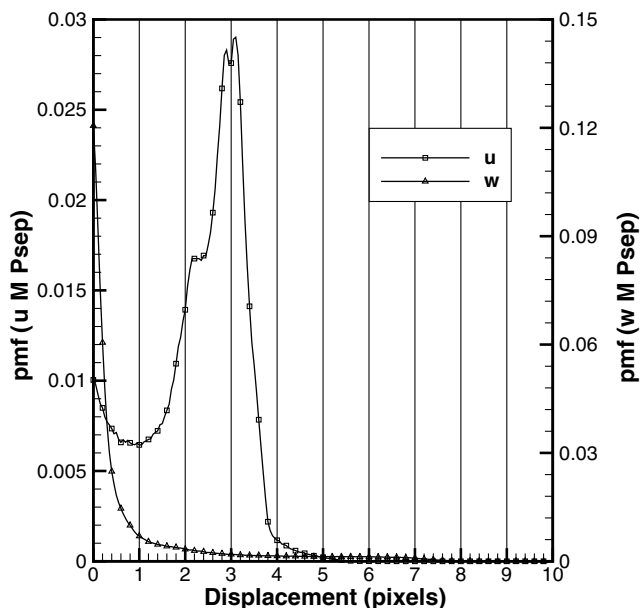


Fig. 18 A representative probability mass function (pmf), or weighted histogram, for the u and w velocity components as measured by the 2-D PIV. There is no indication of pixel locking.

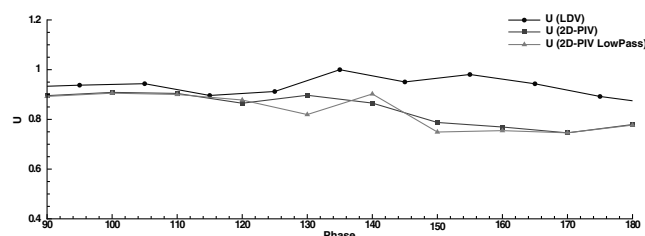


Fig. 19 Effect of low pass filtering the instantaneous PIV data before phase averaging.

located at integral pixel displacements, was not found. The process of removing any cycle-to-cycle variation in the instantaneous PIV velocity fields was also examined. The instantaneous PIV velocity fields were low pass filtered using a large eddy simulation (LES)-style scale decomposition [42]. These filtered fields were then phase averaged and a new U velocity signature constructed. As shown in Fig. 19, the filtering does affect the signature during the jet interaction, leaving the signature unaffected elsewhere. This provides an indication that there is some cycle-to-cycle variation. However, there is still an offset with respect to the LDV data that is not uniform and is greater than the offset during the suction cycle.

Finally, the response of the tracer particles themselves was considered. The flow was seeded with $0.86 \mu\text{m}$ PSL particles for the LDV portion of the experiment and seeded with $3.6 \mu\text{m}$ mineral oil droplets during the PIV portion of the experiment. Both particles have a filterlike effect on the measurements of the actual fluid velocity. It is expected that the larger particles used for the PIV will have a larger response time and therefore act more as a low-pass filter to the velocity fluctuations than do the PSL particles. The PIV interrogation method itself requires multiple particles to participate in the determination of the velocity and acts as an additional filter on the velocity fluctuations that can be measured [43]. This is opposed to the measurement of the velocity of a single particle enforced by the coincidence mode LDV. Whether the observed discrepancies in the velocity signatures are solely due to the different particle dynamics in the two techniques is still not clear. It may be that the large variance in the flow at this part of the cycle affects each of the measurements from the two instruments in fundamentally different ways. However, the LDV data are considered to be the better representation of the actual phase-averaged velocity field, because the LDV has a smaller spatial resolution, $100 \mu\text{m}$ sphere versus $\sim 500 \mu\text{m}$ area, the measurement is always based on the velocity of a single micron sized

particle, and the effect of the averaging bin size was demonstrated to be minimal.

VI. Conclusion

The details of the experimental effort that supports case 2 of the NASA Langley Research Center Workshop on CFD Validation of Synthetic Jets and Turbulent Separation Control have been documented herein. The specification of the boundary conditions, the inflow boundary layer profile and turbulence quantities, the geometry of the tunnel and actuator, and the actuator cavity parameters as a function of phase have been documented and presented. A detailed description of the flowfield data acquired by LDV and PIV has been presented, along with a description of the processing and uncertainty of the measurements that are required to arrive at the phase-averaged data. The level of detail documented here, and in the publicly available data, allows this experimental effort to be used as a useful validation database for flow control simulation.

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